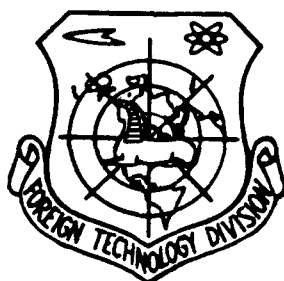


1. The first step in the process is to identify the problem or issue that needs to be addressed. This involves gathering information and understanding the context of the problem.

2

(Selected Pages)

DTIC  
ELECTE  
JAN 16 1992



**92-01381**



Approved for public release;  
Distribution unlimited.



## HUMAN TRANSLATION

FTD-ID(RS)T-1369-90

5 November 1991

"AERODYNAMIC FIGURE CAPTION" (Selected Pages)

English pages: 4

Source: Guoji Hangkong, Nr. 5, 1988,  
pp. 67; 70-71

Country of origin: China

Translated by: SCITRAN  
F33657-84-D-0165

Requester: FTD/TTMM/Moorman

Approved for public release; Distribution unlimited.

THIS TRANSLATION IS A RENDITION OF THE ORIGINAL FOREIGN TEXT WITHOUT ANY ANALYTICAL OR EDITORIAL COMMENT. STATEMENTS OR THEORIES ADVOCATED OR IMPLIED ARE THOSE OF THE SOURCE AND DO NOT NECESSARILY REFLECT THE POSITION OR OPINION OF THE FOREIGN TECHNOLOGY DIVISION.

PREPARED BY:

TRANSLATION DIVISION  
FOREIGN TECHNOLOGY DIVISION  
WPAFB, OHIO

# GRAPHICS DISCLAIMER

All figures, graphics, tables, equations, etc. merged into this translation were extracted from the best quality copy available.

Accession For		
NTIS GRA&I	<input checked="" type="checkbox"/>	
DTIC TAB	<input type="checkbox"/>	
Unannounced	<input type="checkbox"/>	
Justification		
By		
Distribution		
Availability Codes		
Avail and/or		
Dist	Special	
A-1		

(page 1 of the original --)

Figure 2: Blade back flow spectrum for  $\alpha=0^\circ$  and  $I=0.53$ .

Figure 3: Blade bowl flow spectrum for  $\alpha=0^\circ$  and  $I=0.53$ .

Figure 4: Flow spectrum for  $\alpha=10^\circ$  and  $I=0.53$ .

Figure 7: Increase of whirlpool string for  $\alpha=0^\circ$  and  $I$  increases to 1.0.

Figure 8: Jet stream departs from blade back and appearance of whirlpool for  $\alpha=10^\circ$  and  $I=0.60$ .

Figure 9: The adherence of the jet streams of the first and second vent holes with blade back is better when  $\alpha=-9^\circ$  and  $I=0.28$ .

Figure 10: The adherence of the jet stream of the first vent hole with blade back is degraded and whirlpool appears when  $\alpha=-9^\circ$  and  $I=0.60$ .

(page 2 of the original --)

1. Outlook of the test laboratory
2. Flow-conducting sound-damping plate
3. Sound-absorbing wedge
4. Sound insulation room of the exhaust sound-absorbing tube
5. Outlet of exhaust sound-absorbing tube
6. Removable plate-type sound absorber (left side was closed and right side was opened)
7. Exhaust maze

(page 3 of the original --)

1. Outlook of test laboratory
2. Main control room, the data collecting system is on the left and on the right side is three PDP 11/73 computers
3. J-6 fighter awaits testing
4. J-6 fighter in fatigue test

(page 5 of the original --)

1. Combustion Turbine Research Institute
2. full size dual-shaft turbine composite testing equipment
3. single multiple-stage compressor testing equipment
4. high altitude station gas supply plant
5. high altitude simulation testing cabin
6. supersonic compressor plane-blade grid wind tunnel
7. supersonic air inlet simulation wind tunnel
8. high altitude station exhaust cooling system

(page 6 of the original --)

Figure 2: Y-7 assembly building of Xian Aircraft Company

Figure 3: Overlay skin stretching machine

Figure 4: Microwave testing dark room

(page 7 of the original --)

Figure 1: Digital riveting machine

(page 8 of the original --)

Figure 2: DNC technology development

Figure 3: Scanning electron microscope

Figure 4: Large-scale hot pressing canister

Figure 5: Vacuum electrical furnace

Figure 6: Air-circulation quenching furnace

(page 9 of the original --)

Figure 1: Main laboratory of computer center

Figure 2: CAD design in the design center

(page 10 of the original --)

Figure 1: Full-scale measuring machine

(page 11 of the original --)

Figure 2: penetration detecting plant (interior view)

Figure 3: Circularity measuring device for high-precision parts

Figure 4: Electrostatic spray penetration detecting machine

(page 12 of the original --)

Figure 2: System dynamic simulation in the engine fuel system  
control laboratory

(page 13 of the original --)

1. Low-speed wind tunnel simulation test
2. Full craft turbine illustration by the 7-hole probe
3. High-speed wind tunnel open-hole wall

**DISTRIBUTION LIST**  
-----

**DISTRIBUTION DIRECT TO RECIPIENT**  
-----

<b>ORGANIZATION</b> -----	<b>MICROFICHE</b> -----
B085 DIA/RTS-2FI	1
C509 BALL0C509 BALLISTIC RES LAB	1
C510 R&T LABS/AVEADCOM	1
C513 ARRADCOM	1
C535 AVRADCOM/TSARCOM	1
C539 TRASANA	1
Q592 FSTC	4
Q619 MSIC REDSTONE	1
Q008 NTIC	1
Q043 AFMIC-IS	1
E051 HQ USAF/INET	1
E404 AEDC/DOF	1
E408 AFWL	1
E410 ASDTC/IN	1
E411 ASD/FTD/TTIA	1
E429 SD/IND	1
P005 DOE/ISA/DDI	1
P050 CIA/OCR/ADD/SD	2
1051 AFIT/LDE	1
CCV	1
PO90 NSA/CDB	1
2206 FSL	1

Microfiche Nbr: FTD91C000675  
FTD-ID(RS)T-1369-90